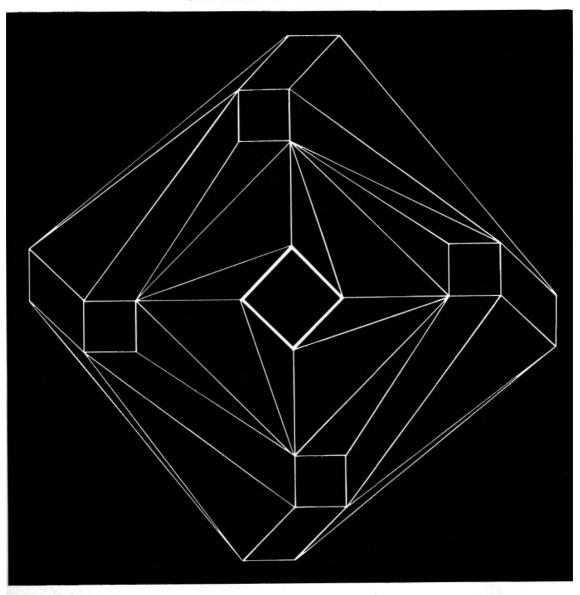
# **THEORETICAL** AND APPLIED MECHANICS

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# Enhancement of Engine Exhaust Flow Attachment for USB Configurations

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One of the difficulties with USB-STOL aircraft relates to the attachment of the engine exhaust flow to the wing and extended USB flap surfaces during low-speed operations.

We have developed a device, side fences, as one of the solutions for enhancing Coanda flow attachment and bring about efficient powered-lift as applied to the USB-STOL propulsive lift system. In the present paper, we attempt to summarize the the principal aerodynamic characteristics that affect the powered lift augmentation for USB configurations, as ascertained by a wind tunnel experiment together with surveys of the engine exhaust flow field over the wing and USB flap surfaces using the 8%-scale semispan model of NAL Quiet STOL Research Aircraft, with special emphasis on the basic properties of the mechanisms for the enhancement of exhaust flow attachment by side fences. Results were compared with those attained either by using the system without special devices or by using vortex-generating vanes.

#### Nomenclature

$AR_{\epsilon}$	: Aspect ratio defined as nozzle area A divided by	q	: Freestream dynamic pressure, $=0.5 ho U_{\infty}^2$
	square of nozzle maximum	S	: Wing area
	height $D_1 = A/D^2$	T	: Static thrust force based
$C_{j}$	: Thrust coefficient defined		upon engine calibrations
	as engine thrust T divided		with flap removed,
	by qS		$=\sqrt{F_N^2+F_A^2}$
$C_L$	: Lift coefficient defined as	$\alpha$	: Angle of attack, deg.
	lift force divided by free-	$\Delta P$	: Static pressure difference
	stream dynamic pressure		relative to ambient pres-
	q times wing area S,		sure $P_0$
	= Lift/ $qS$	WSTA	: Spanwise position on USB
$C_{p}$	: pressure coefficient defined		flap, wing station
	as $\Delta P$ divided by jet mean	$\delta_f$	: USB flap deflection angle,
	dynamic pressure at noz-		deg.
	zle exit, = $\Delta P/0.5 \rho U_{\rm jet}^2$	$\delta_{j}$	: Jet turning angle, deg.
$F_A$	: Axial force	$\eta_j$	: Jet turning efficiency
$F_N$	: Vertical force	γ	: Flight path angle, deg.

## I. INTRODUCTION

The National Aerospace Laboratory (NAL) of Japan has since 1977 been conducting